ACOUSTIC TESTING OF A SUPERSONIC TIP SPEED FAN WITH ACOUSTIC TREATMENT AND ROTOR CASING SLOTS

(Quiet Engine Program Scale Model Fan C)

by

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GENERAL ELECTRIC COMPANY



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Acoustic tests were conducted on a high tip speed (1550 ft/sec, 472.44 m/sec) single-stage fan with varying amounts of wall acoustic treatment and with circumferential slots over the rotor blade tips. The slots were also tested with acoustic treatment placed behind the slots. The wall treatment results showed that the inlet treatment is more effective at high fan speeds and att duct treatment is more effective at low fan speeds. Maximum PNL's on a 200-foot (60.96 m) sideling showed the untreated slots to have increased the rear radiated noise at approach. However, when the treatment was added to the slots the inlet radiated noise was decreased, resulting in little change relative to the solid casing on an EPNL basis.			the rotor lots. The n speeds and t (60.96 m) proach.	
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I. SUMMARY

An acoustic test program was conducted on a 36 inch (91.44 cm) in diameter scale model (0.527 linear scale) of the outer flowpath of the high tip speed, single-stage Fan C of the NASA Quiet Engine Program. The fan had a design point pressure ratio of 1.6 at a tip speed of 1550 ft/sec (472.44 m/sec). Six configurations were tested with varying amounts of acoustic treatment and with slots above the rotor blade tips.

Table I summarizes the front and rear quadrant maximum 200-fcot (60.96 m) sideline PNL's for each configuration. Full nacelle treatment (typical of present day quiet engines, i.e., CF6) reduced the front maximum noise by 8.3 PNdB at takeoff power (90% corrected fan speed). At approach power (57.5% corrected fan speed) the front reduction was 7.2 PNdB. The aft duct suppression was more effective at approach than at takeoff.

A massive aft suppressor was added to isolate the inlet radiated noise. The results show that the noise in the front quadrant is totally inlet radiated even without the suppressor.

Finally, a circumferentially slotted tip casing was employed in order to determine the acoustic effect of slots designed to improve fan stall marigin. Acoustic treatment was also placed behind the slots to determine if any suppression could be obtained. As Table I shows the slots increased the aft quadrant noise over the frame treatment solid casing levels, particularly at approach. The addition of the treatment reduced the levels slightly.

The noise levels extrapolated to level flight produced results similar to Table I for the treated configurations. In the case of the slotted tip casing, however, the reduction in front radiated noise with the treated slots at approach (2.1 PNdB relative to frame treatment in Table I) results in little EPNdB increase. Table II summarizes the result of EPNL extrapolation.

Table I. Maximum 200-Foot (60.96 m) Sideline PNL (Full Scale, Nominal Operating Line).

	Takeof f		Λpproach	
Configuration	Aft	Front	Aft	Front
Untreated	119.7	123.1	105.8	105.3
Frame Treatment	117.2	120.4	102	102.6
Full Nacelle Treatment	112	114.8	96.2	98.1
Massive Aft Suppression	110.5	114.9	94.5	98.3
L'otted Tip (Untreated)*	118	120	104.2	101.4
Slotted Tip (Treated)*	117.3	119.8	103.7	100.5

^{*}Slotted configurations include Frame Treatment. The Slotted Tip (Treated) implies treatment behind the slots.

Table II. Level Flyover* EPNL, Fan and Predicted Core Jet, Single Engine.

Configuration	Takeoff 1000' (304.8 m) Alt	Approach 370' (112.8 m) Alt
Untreated	105.2	96.7
Frame Treatment	101.2	93.2
Full Nacelle Treatment	95.4	89.8
Slotted Tip (Untreated)	101.7	94.9
Slotted Tip (Treated)	100.9	93.6

^{*} Flight Mach No. = 0.25

II. INTRODUCTION

From an engine system viewpoint, a high tip speed fan is desirable when high specific fan thrust is required. This is because the high speed fan can produce the needed pressure rise in a single stage and with fewer low pressure turbine stages. However, the high tip speed fan poses a unique noise problem in the form of shock or multiple pure tone (MPT) noise. Typical spectral characteristics of this type of noise are shown in Figure 1.

Recognizing this fundamental difference between the noise signatures of high and low tip speed lans, two basic fan designs were evolved for the Quiet Engine Program.^{1,2} One of the low speed fans (two were designed) and the high tip speed fan were not only built in full scale (for 22,000 pound, 97,900 newtons, thrust engines), but also partially modeled in approximately one-half scale size.³ In these scale models only the bypass flow portion of the fan flowpath was modeled.

This report is concerned with the high tip speed fan scale model - Fan C. Six configurations of this fan are considered:

- 1. No acoustic treatment
- 2. Fan frame treatment
- 3. Full nacelle wall treatment
- 4. Full nacelle wall treatment with a massive aft suppressor
- 5. Slotted tip casing (untreated)
- 6. Slotted tip casing (treated)

(A detailed description of each configuration is contained in the next section of this report.)

The untreated vehicle serves as a baseline for the test series. Fan frame treatment was an intermediate treatment configuration which was defined as that treatment which would normally be supplied by the engine manufacturer, as opposed to treatment applied to the inlet and exhaust nacelles which are usually associated with the installed engine. For the next level of suppression, the nacelle treatment was added. This treatment is as extensive as might be found in the newest generation of high bypass turbofan engines.

The nacelle treatment with the massive aft suppressor was utilized to determine the extent of the inlet radiation in this configuration. This suppressor was designed to effectively eliminate the aft generated noise radiated to the front quadrant.

Slots above the rotor had been demonstrated to improve fan stall margin. However, the effect these slots would have on a fan noise generation were unknown. In addition, since the slots were above the rotor, it was determined that acoustic treatment could be placed behind the slots and possibly afford increased noise suppression. Thus, the slots were run with and without the treatment behind the slots.

III. VEHICLE AND TEST FACILITY DESCRIPTION

The relationship of the scale model to the full-scale fan is shown in Table III and Figure 2. Except for the radius ratio difference all dimensions and aerodynamic parameters obey the usual scaling rules. The radius ratio divergence is shown in Figure 2 by the dashed lines. Motive power for the fan is through a front shaft as shown in the photograph in Figure 3.

Figure 4 shows the sound field and the acoustically enclosed drive engine. Microphones are placed every 10 degrees from 30 degrees to the inlet axis around to 160 degrees. The microphone arc is 100 feet (30.48 m) centered at the fan inlet hub; while the field between the microphones and the vehicle is rovered with asphalt.

Figure 2 also shows the location of the acoustic treatment for both the frame and nacelle treatment. The full nacelle includes the frame and nacelle treatments. In each segment the treatment is made up of 1/2 inch (1.3 cm) of polyurethene foam covered with a perforated plate having a porosity of 22-1/2%. The holes are 1/16 inch (0.2 cm) in diameter. For untreated configurations, these treated areas were made acoustically "hard" by covering them with metal impregnated tape.

Also shown in Figure 2 is the massive aft suppressor. This piece of hardware was employed to help isolate the front end noise by effectively eliminating the rear radiated noise. The suppressor is 43 inches (109.22 cm) long and contains a 20-inch (50.8 cm) splitter. All surfaces are treated with polyurethene foam covered with a perforated face plate. The suppressor was designed so as not to increase the backpressure on the fan by increasing the flow cross sectional area to account for the splitter and boundary layer buildup.

A cross section of the slotted tip casing is shown in Figure 5. The number of circumferential slots was determined from previous aerodynamic testing 4 as providing the best of increased stall margin and smallest efficiency loss. From the acoustic viewpoint these slots may also be better since, as opposed to axial or oblique slots, the circumferential slots do not present a periodic geometry change relative to the rotor blades.

Outside of the slots the single-degree-of-freedom acoustic treatment can be seen. This treatment is coupled to the slots by holes drilled through the bottoms of the slots. The surface porosity is approximately 11%. This design results in a Helmholtz resonance of about 2500 Hz and a 1/4 wave resonance of 6700 Hz.

Table III. Scale Model and Full-Scale Fan Design Parameters.

Parameter	Scale Model	Fall Scare
Diameter, in. (cm)	36.0 (91.44)	68.3 (173.48)
Design Tip Speed, ft/sec (M/sec)	1550 (472.44)	1550 (472.44)
Design Fressure Ravio	1.6	1.6
Weight Flow, 1.b _m /sec (kg/sec)	187 (84.9)	850 (385.5)
Radius Ratio	.57	.36
Number of Blades	26	26
Blade-to-Vane Spacing, chords	2.0	2.0
Number of Vanes	60	60

IV. TEST PROGRAM AND DATA ANALYSIS

Each configuration was setup on the test facility and run through its operating speed range (50 to 100%) in sufficient steps (10 or 11 points per operating line) to fully define part power characteristics. In addition the untreated, frame treatment, and full nacelle configurations were run with exhaust nozzle areas which were smaller (6%) and larger (16%) than the nominal nozzle.

Noise data were recorded at each microphone for two minutes at each corrected speed point. This procedure was repeated once so that all data are the average of two points unless otherwise stated. The recorded data were processed through a General Radio 1/3-octave band analyzer utilizing a 32-second averaging time. Standard correction factors were applied to bring these data to a standard day of 59° F and 70% relative humidity.

The data represented, of course, the noise signature of the scale model. In order to better assess the PNL results, these data were scaled to full scale by adding a factor of 10 log of the ratio of the full scale to scale model weight flows to all the data and shifting the frequency down by the ratio of the blade passing frequencies of the full scale and scale model. Unless otherwise noted, all the data presented in this report are full scale.

Also of interest are extrapolation of these data to flight. The flight noise calculation was enhanced by adding a predicted core jet and accounting for the relative velocity effect. Core jet noise and relative velocity effects were predicted according to published SAE practices⁶.

V. NACELLE TREATMENT

A. Static 200-Foot (60.96 m) Sideline Data

The extent of the nacelle treatment is shown in Figure 2 along with frame treatment and massive aft suppressor cross sections.

Figure 6 shows the PNL at takeoff fan speed (90% corrected). Characteristically, the high speed fan has its peak noise in the front quadrant — at 70 degrees. The addition of frame treatment reduces the front peak by 2.7 PNdB and the rear peak by 2.5 (comparing 120 degrees untreated to 130 degrees treated). Further addition of treatment to the full nacelle reduces the front noise by 8.3 PNdB and the rear noise by 7.7 PNdB relative to the untreated case. In each configuration, however, the fan remains front dominated.

Finally, the addition of the massive aft suppressor shows an insignificant change in the noise level. This signifies that the front radiated noise dominates the PNL's.

Figures 7 and 8 are spectral comparisons at, respectively, 70 and 120 degrees for the takeoff condition. The spectral peak at 500 Hz is controlled by multiple pure tones. At 70 degrees these tones dominate the spectrum; while the 120 degree data show the blade passing frequency (2 KHz) to be at about the same level as the 500 Hz band. Addition of the frame treatment reduced the 500 Hz noise by 8.5 dB at 70 degrees and the rest of the high frequency spectrum by lesser amounts. At 120 degrees both dominating bands (500 and 2000 Hz) were reduced by about 4 dB.

Extending the inlet treatment by 10.5 inches (26.67 cm) had a profound effect on the inlet noise. The MPT's have been eliminated as major contributors to the PNL, although examination of narrowbands does show some MPT content, particularly around 400 Hz. If thicker treatment had been added to the inlet (the existing treatment was 1/2 in. thick) these MPT's probably would have been removed. At 120 degrees, the MPT's have been eliminated indicating that the inlet treatment weakened the MPT's to the extent that they no longer radiate into the aft quadrant. The BPF has been reduced by about 14.5 dB.

The massive aft suppressor increased suppression at frequencies above 3150 Hz. The noise remaining is believed to be solely fan jet noise and flow scrubbing noise from inside the suppressor.

Figures 9-11 show the PNL directivities for, respectively, 84, 72, and 57.5% (approach) fan speeds. Generally, the unsuppressed data move from front dominant to slightly rear dominant at approach. The front noise suppression at approach due to the frame treatment is about 2.7 PNdB (comparing 70 degrees untreated and 60 degrees treated) while the full inlet treatment results in a 7.2 PNdB (70 degrees untreated relative to 40 degrees treated) reduction. The rear noise reduction is slightly greater - 3.8 PNdB for the frame and 9.6 PNdB for the full treatment.

Spectral comparisons at approach for 50, 70, and 120 degrees are shown, in Figures 12, 13, and 14 respectively. The BPF is in the 1250 Hz band. At 70 degrees, the tone-controlled bands show the greatest reduction. At 120 degrees, the suppression for both frame and full treatment extends over all of the high frequency spectrum with the maximum frame suppression being 7.5 dB at 1650 Hz and the maximum full treatment suppression being 16.4 dB at 2 KHz. The persistence of the BPF level at 1250 Hz when the massive aft suppressor is added is not fully understood. However, two possibilities exist; either the noise came from the inlet or there is a flanking path through the vehicle's structure.

Figures 15 and 16 show the progression of front and aft 200-foot (60.96 m) sideline maximum PNL's for each configuration versus corrected fan speed. The untreated front maximum data show a break upward at about 80% speed and a downward turn at 95% speed. The increase at 80% signifies the onset of MPT radiation sufficient to effect the PNL; while the break at 90-95% speed represents the full swallowing of the bow shock wave. When treatment is added, however, the curves begin to smooth out. With full treatment the curve tends to start leveling at 90% speed. The aft maxima are generally smooth with suppression levels being nearly constant over the entire speed range.

Figures 17 through 20 contain maximum PNL information for the frame and full treatment configurations when two other nozzles were employed. One nozzle area was smaller than nominal by 6% and the other was larger than nominal area by 16%. When frame treatment was employed, the large nozzle resulted in slightly higher noise at constant thrust (Figures 17 and 18); while the small and nominal nozzle data were nearly coincident. This same trend was observed when full treatment was used; particularly at the front maximum angle. Figure 21 shows the PNL directivity at approximately takeoff thrust for the large and nominal nozzles. The first thing to be noted is that in order to get the same thrust with the large nozzle, a fan speed increase in excess of 95% speed is required. As a meneral rule the higher speed tends to drive the front noise up. This expending the increased front noise with the large nozzle. The 60 degree spectral comparison is shown in Figure 22. The noise increase is largely concentrated around the BPF (2 KHz).

B. Flight Noise

In order to better assess the noise reduction obtained with the frame and full treatment, the takeoff and approach results were "flown" on a level flight path at 1000 feet (304.8 m) for takeoff and 370 feet (112.776 m) for approach. Also included was a prediction of the core jet noise based on Engine C cycle data and the method of Reference 6.

Figures 23 and 24 show the PNL and PNLT for takeoff fan speed. The loise reductions for frame and full treatment are similar to the PNL reduction at the 200-foot (60.96 m) sideline (Figure 6). When the tone correction is in force, Figure 24, the full treatment reductions increase; particularly at the aft maximum angles.

At approach fan speed, Figures 25 and 26, the PNL and PNLT reductions are again similar to the PNL reduction at static conditions (Figure 11).

Table IV is a summary of the EPNL values derived from Figures 24 and 26. Nearly 10 EPNdB reduction is obtained at takeoff with full nacelle treatment. The approach reduction, where MPT's are not present at the source, is less - 6.9 EPNdb.

C. Comparison to Full-Scale Fan Data

The scale model fan was a 0.527 linear scale of the outer flowpath of the full-scale Engine C fan (see Section III-B). Engine C was tested by General Electric and the full-scale Fan C was tested as a component by NASA. The data presented in this report has been scaled to the full-scale size (see Section IV-C).

Comparisons of these three sets of data must be interpreted in terms of the installation and site conditions under which each was tested. The engine was run over a gravel field while both the fans were run over asphalt surfaces. Both fan component vehicles were driven by front shafts which required pedestal bearings, while the engine's inlet was relatively clean.

The asphalt to gravel difference has been derived by testing the scale model fan over both surfaces in the frame treated condition. The data obtained were averaged over the speed range and around the microphone arc and finally a smooth line was curve-fitted through the result. The line is shown in Figure 27.

Differences caused by the front shaft drives have not been quantified; however, as would be expected, the differences are concentrated in the front quadrant.

There are also some vehicle differences. The scale model is only the outer panel of the fan while the engine and full-scale fan contain the full-span blade. As described earlier, the scale model has Scottfelt treatment while the full-scale fan and engine use a multiple-degree-of-freedom resonater treatment. The differences in acoustic data attributed to these two characteristics have not been quantitatively established.

Figure 28 contains the PNL for the three vehicles at takeoff fan speed. At the rear angles the agreement is very good. Forward of 60 degrees, the data spread out with the scale model data being the lowest and the full-scale fan data the highest. The lower level of the scale model is probably due to "blocki g" of the forward radiated noise by the drive engine housing. The higher level of full-scale fan noise has been generally attributed to higher inlet turbulence levels generated by air flowing over the pedestal bearing.

Table IV. Level Flyover* EPNL, Fan and Predicted Core Jet, Single Engine.

Configuration	Takeoff 1000' (304.8 m) Alt	Approach 370' (112.8 m) Alt
Untreated	105.2	96.7
Frame Treatment	101.2	93.2
Full Nacelle Treatment	95.4	89.8

* Flight Mach No. = 0.25

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The 70 and 110 degree spectra are shown in Figures 29 and 30. Considering the highly tone-controlled nature of the spectrum (multiple pure tones) the agreement is quite good. Some of the lower level high frequency noise from the engine is probably attributable to the surface differences (Figure 27). The higher engine noise at low frequencies (below 160 Hz) is due to the engine's core jet. At 110 degrees there is more spread at low frequencies. Some of this may be connected with the surface reflection conditions which have more of an effect on low frequency noise than high frequency noise. The level of the blade passing frequency at 2 KHz is almost the same for all three vehicles. There is no apparent explanation for the lower level of scale model noise at high frequencies.

The approach PNL's, Figure 31, show considerable difference; particularly in the rear quadrant. Spectral comparisons are shown in Figures 32 and 33. At 120 degrees, there is a large spread at frequencies above the second harmonic (2.5 KHz). The difference between the engine and the fans was expected since it is in this region that the engine's turbine noise is detected. At the BPF (1250 Hz) agreement was good. Differences at frequencies below the BPF are not easily explained; however, jet noise and reflection pattern changes probably play a part.

Although differences exist between these vehicles, each by itself provides a reliable method of examining the characteristics of a given fan under varying conditions of acoustic treatment.

D. Aerodynamic Performance

The aerodynamic performance was assumed to remain unchanged for each of the treated configurations. Figure 34 is the performance map for the scale model.

VII. SLOTTED TIP CASING

A. Static 200-Foot (60.96 m) Sideline Data

As described in Section III-B, the Fan C rotor was tested with slots over the rotor with and without a single-degree-of-freedom resonator behind the slots.

Figure 35 shows the takeoff PNL directivity at 200 feet (60.96 m). The front angles show very little variation among the three configurations. At the rear angles, particularly at 120 degrees, there appears to be a hierarchy of slots (untreated), slots (treated), and frame (solid casing) in order of decreasing noise level.

Spectral comparisons at takeoff are shown in Figures 36 and 37. The 120 degree spectra indicate a tendency for the slotted data to be noisier than the frame treatment results at frequencies above the BPF (2 KHz).

Figures 38, 39, and 40 contain the 200-foot (60.96 m) sideline PNL for, respectively, 84%, 72%, and approach (57.5%) fan speed. Generally the slotted data is falling above the frame treatment results in the rear quadrant. The difference between the treated and untreated slots is negligible at the rear angles. However, Figure 40 at 60 degrees shows a noticeable decrease in the slotted (treated) level. Figure 40 also indicates that the slotted results are about 2 PNdB higher than frame treatment at 130 degrees. The spectral comparisons, Figures 41 and 42, show a general increase in high frequency noise at 120 degrees. At 70 degrees, there is a lower BPF with the slotted casing with treatment.

Figures 43 and 44 show, respectively, the progression of front and aft maximum 200-foot (60.96 m) sideline PNL with corrected fan speed. In front at low speeds the slotted casings are quieter than the solid casing levels. At higher speeds (above 73%) the opposite is true. The rear noise shows the slotted data roisier than the solid casing at all speeds with the greatest difference at lower speeds.

B. Flight Noise

Figures 45 through 48 show the noise levels extrapolated to level flight conditions. (A core jet has been added along with relative velocity effects⁶.) Generally, the trends are the same as for the ground static data. A notable exception is at approach fan speed, Figure 48, where the PNLT directivity reveals a 4 dB increase in the rear radiated aft maximum noise with the untreated slot and a 3.2 dB decrease with the treated slot at 60 degrees.

Table V summarizes the EPNL results derived from these PNLT histories. The changes at takeoff are small. At approach, however, the untreated slot data show an increase of 1.7 EPNdB. The treated slot data show most of this difference to have disappeared. This is mainly due to a decrease in the front maximum noise when the treatment was added.

Table V. Level Flyover* EPNL, Fan and Predicted Core Jet, Single Engine.

Configuration	Takeoff 1000' (304.8 m) Alt	Approach 370' (112.8 m) Alt
Frame Treatment (Solid Casing)	101.2	93.2
Slotted Casing (Untreated)	101.7	94.9
Slotted Casing (Treated)	100.9	93.6

Flight Mach No. = 0.25

C. Aerodynamic Performance

Along with the acoustic data, nominal nozzle performance data was taken with the slots. Figure 49 shows a comparison of the solid casing and the slotted casing on the performance map. Within the accuracy of the data there is no discernible change. Figure 50 contains the radial distribution of pressure rise, temperature rise, and the resulting efficiency at takeoff fan speed. Again the changes are small, although there are indications that the slotted casing is higher in efficiency.

Finally, Figure 51 shows the average efficiency trends with speed for the solid and slotted casings. The line faired through the solid casing data indicates that the slotted casing data lie largely above the line, particularly at 90% speed.

VII. CONCLUSIONS

- 1. With full nacelle treatment, the inlet suppression is more effective at takeoff than at approach and the aft duct treatment is more effective at approach than at takeoff.
- 2. The slots above the rotor increased the aft radiated noise; particularly at low power settings.
- 3. Including a single-degree-of-freedom acoustic treatment behind the slots reduced front noise levels resulting in a slotted configuration which shows little difference from a solid casing on an EPNL basis.
- 4. The slotted fan casing appeared to enhance the fan efficiency about one percent at corrected speeds near design.

Appendix A - Illustrations

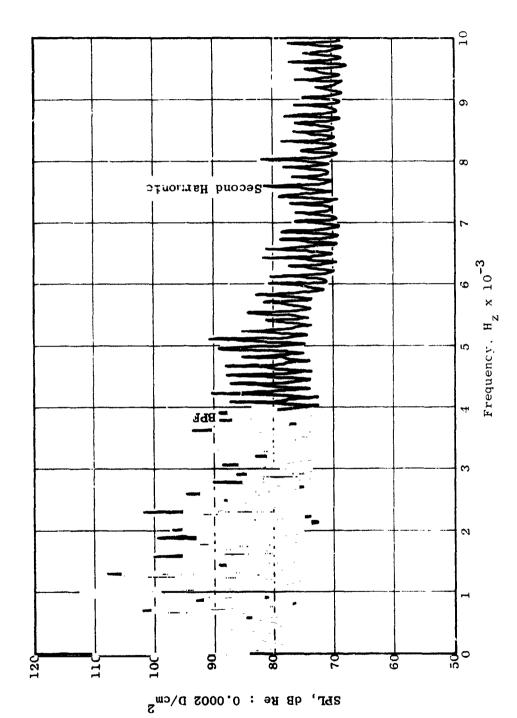


Figure 1, Typical Spectral Characteristics of a High Speed Fan.

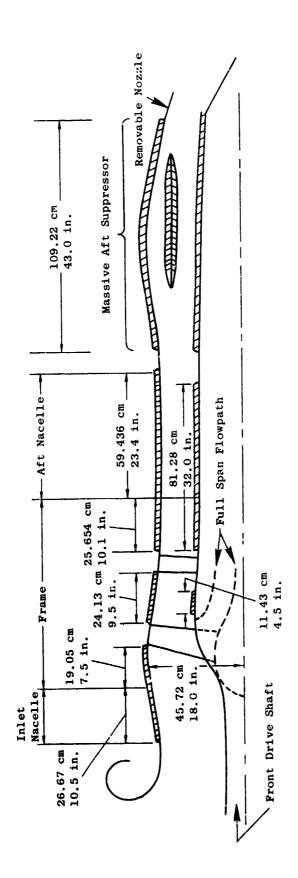


Figure 2. Fan Vehicle Cross Section.

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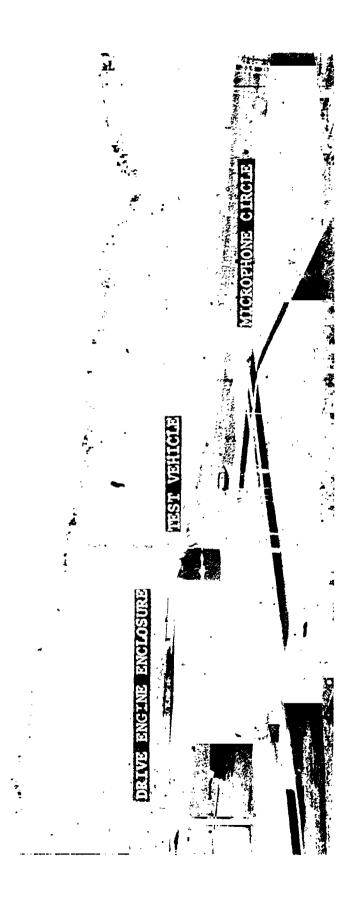


Figure 4. Fan Test Facility.

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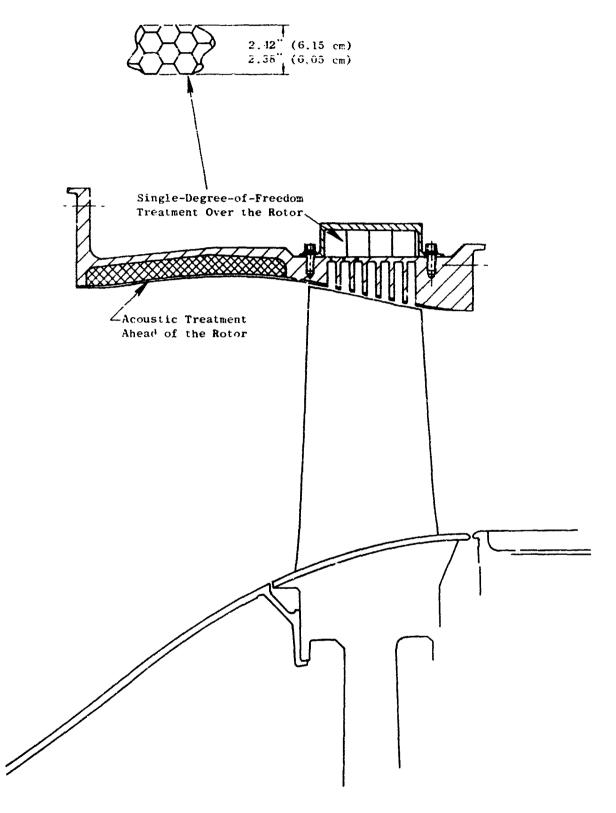
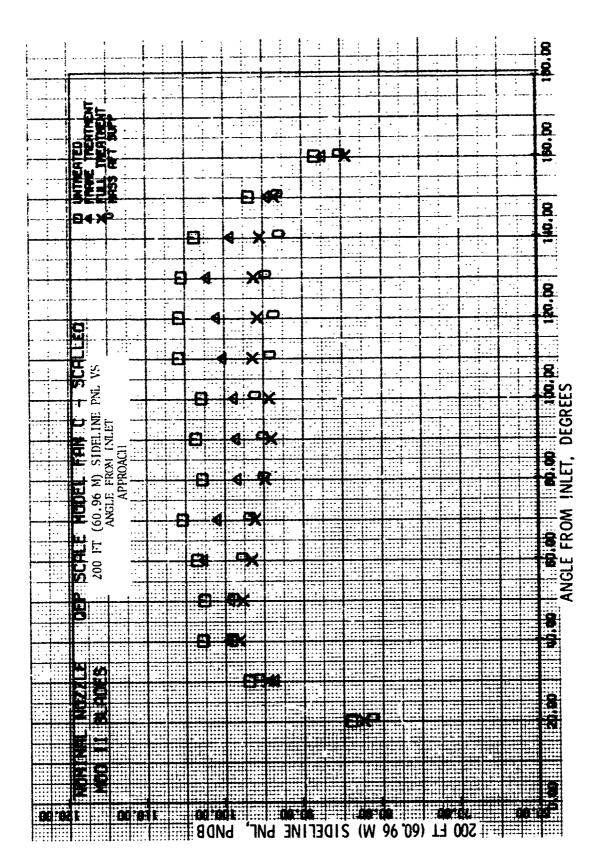
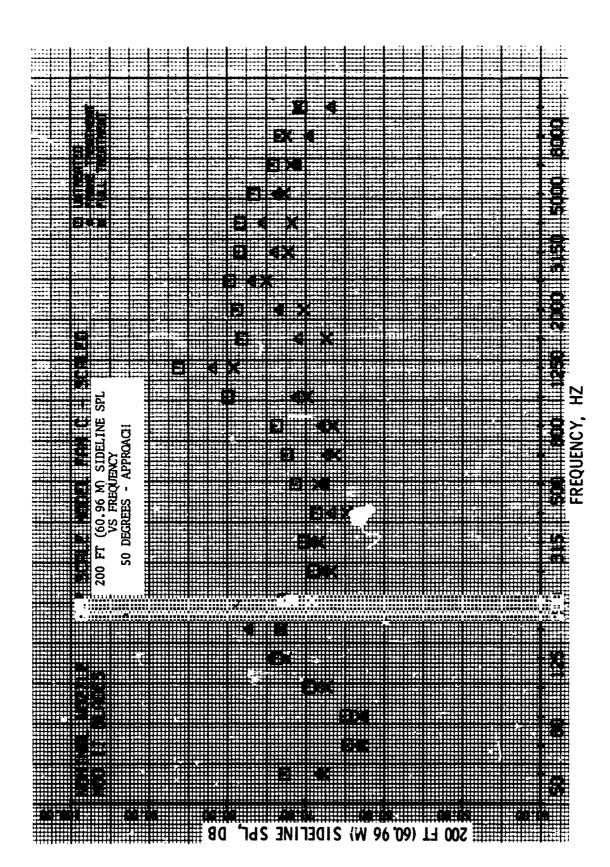


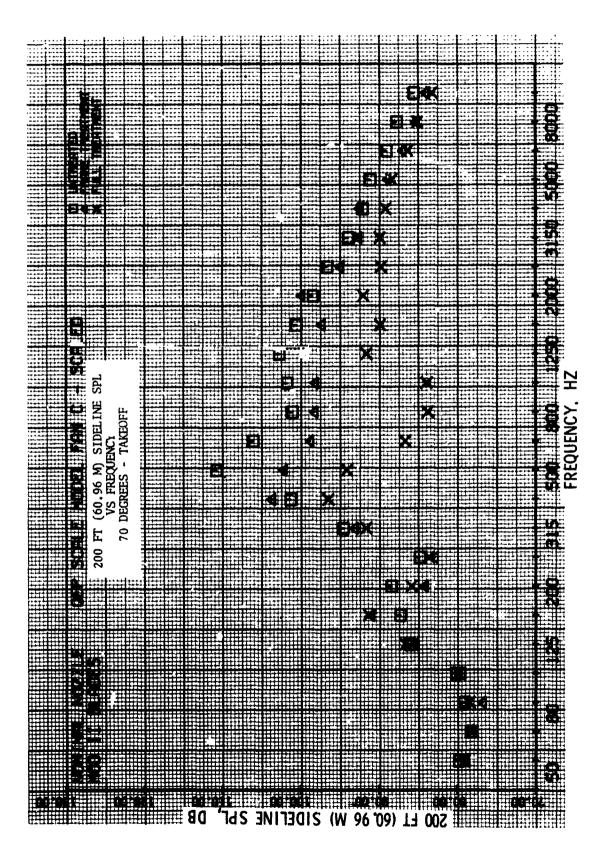
Figure 5. Cross Section of Slotted Tip Casing.



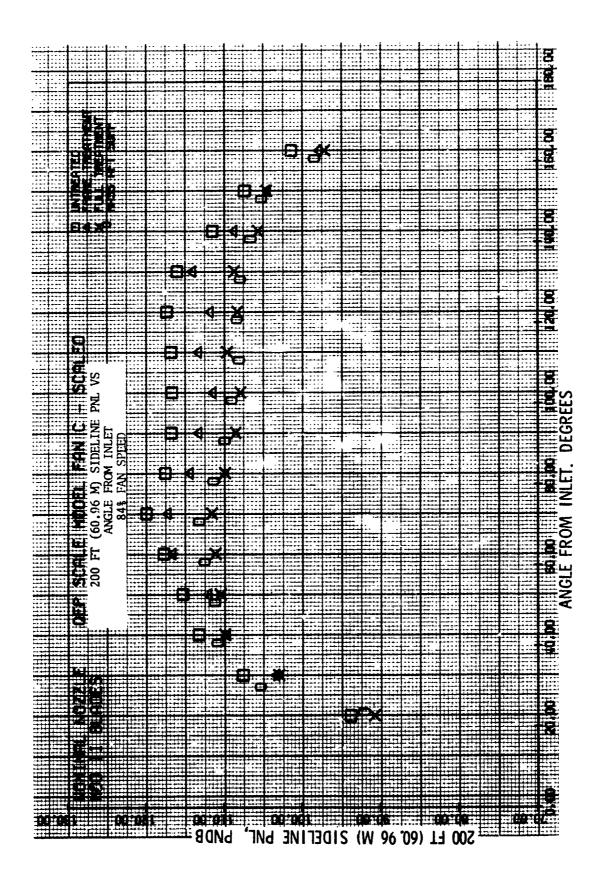
200-ft (60.96 m) Sideline PNL Vs. Angle from Intet, Takeoff. .9 Figure



200-ft (60.96 m) Sideline SPL Vs. Frequency, 70°, Takcoff. Figure 7.

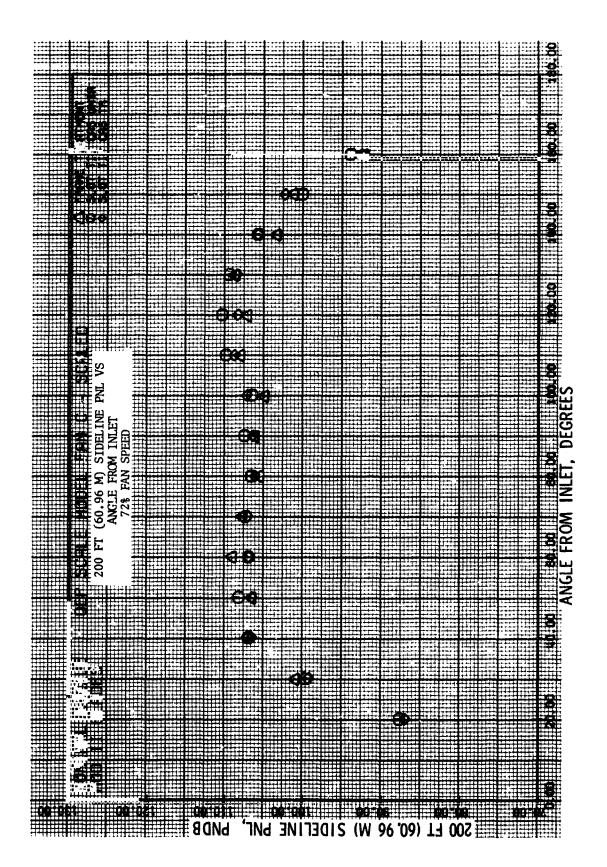


200-ft (60.96 m) Sideline SPL Vs. Frequency, 120°, Takeoff. Figure 8.

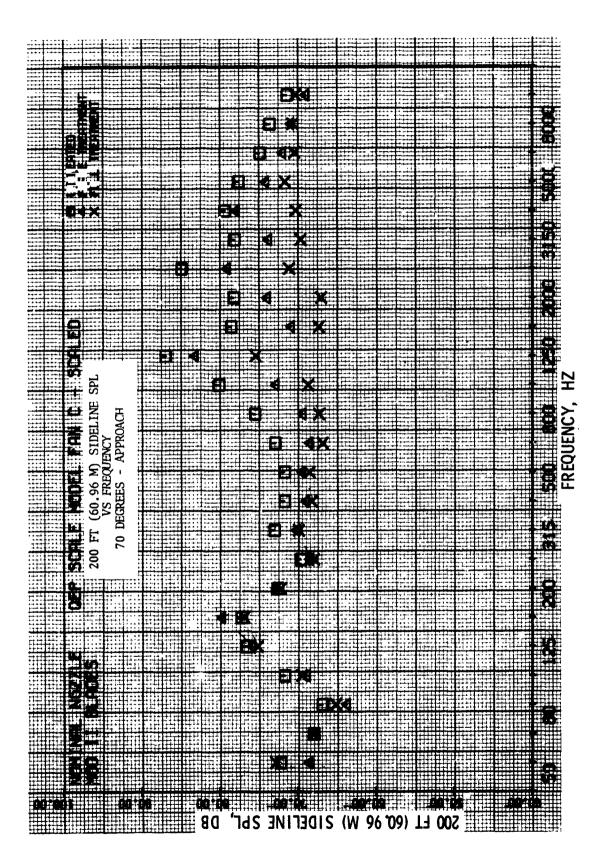


200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 84% Fan Speed. Figure 9.

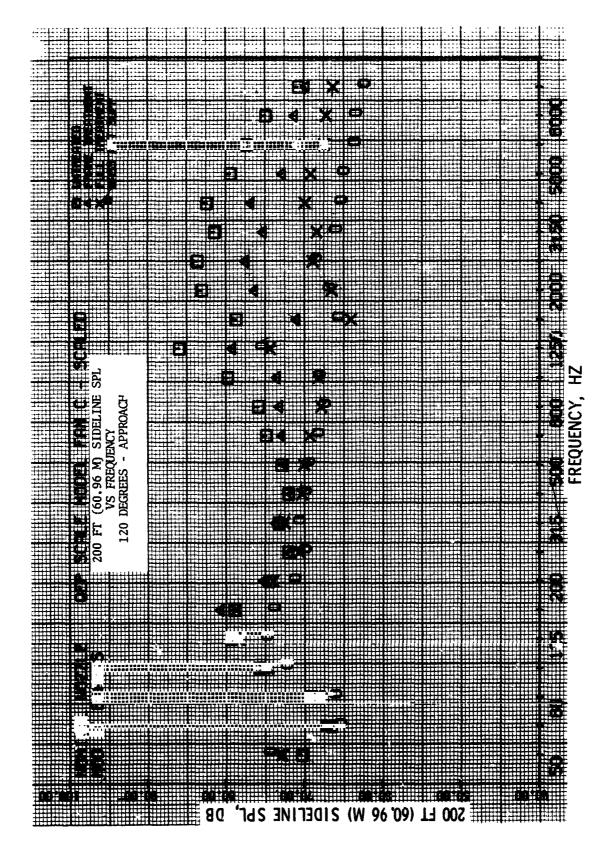
Figure 10, 200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet, 72% Fan Speed.



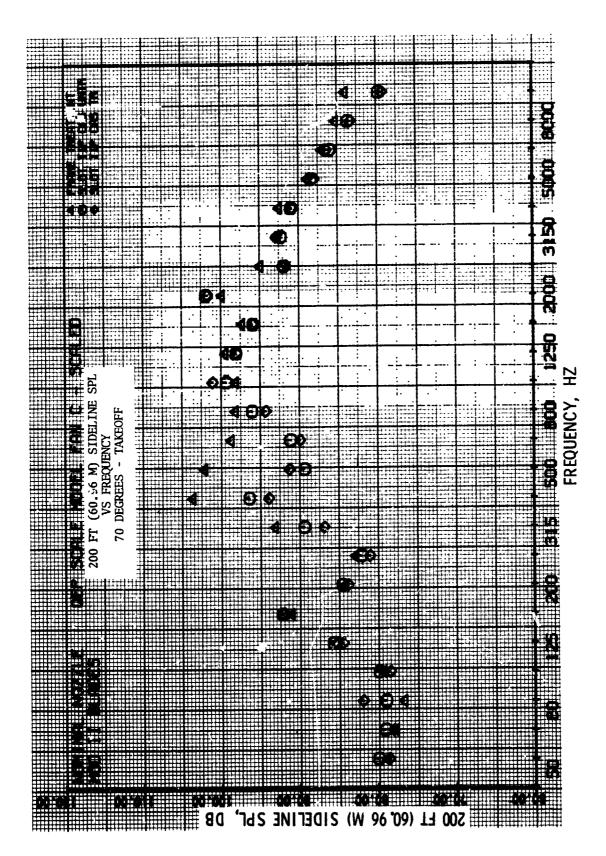
Sideline PNL Vs. Angle from Inlet, Approach. 200-ft (60.96 m)



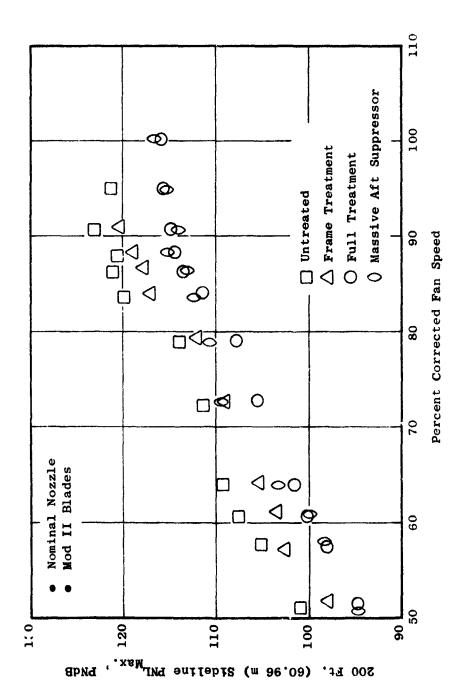
200-ft (60.96 m) Sideline SPL Vs. Frequency, 50° , Approach. Figure 12.



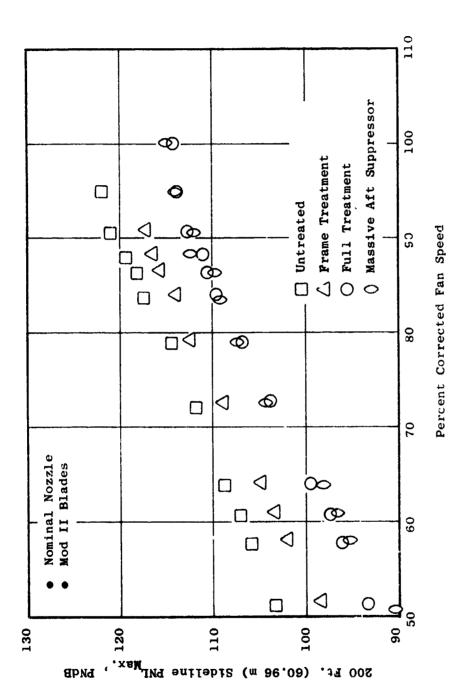
200-ft (60.96 m) Sideline SPL Vs. Frequency, 70°, Approach. Figure 13.



200-ft (60.96 m) Sideline SPL Vs. Frequency, 120°, Approach.

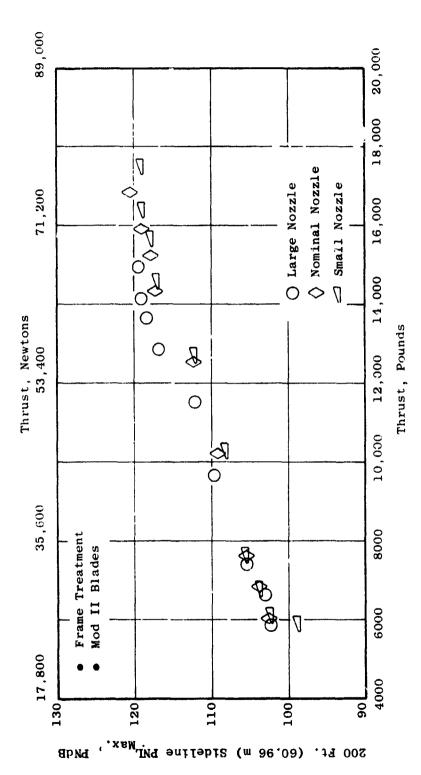


200-(t (60.96 m) Sideline Front Maximum PNL Vs. Corrected Fan Speed. Figure 15.

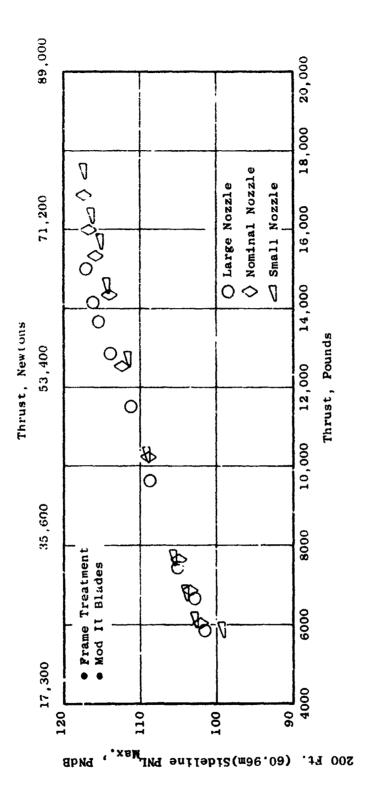


200-ft (60.95 m) Sideline Aft Maximum PNL Vs. Corrected Fan Speci.

Figure 16.

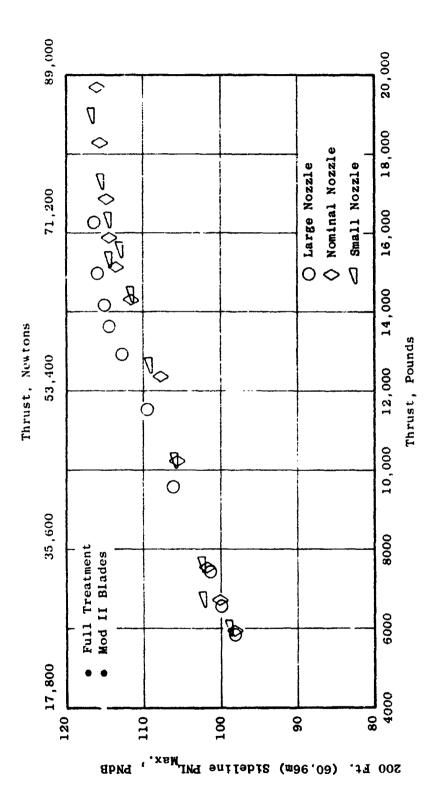


200-ft (60.96 m) Sideline Front Maximum PML Vs. Thrust, Frame Treatment Configuration. Figure 17.

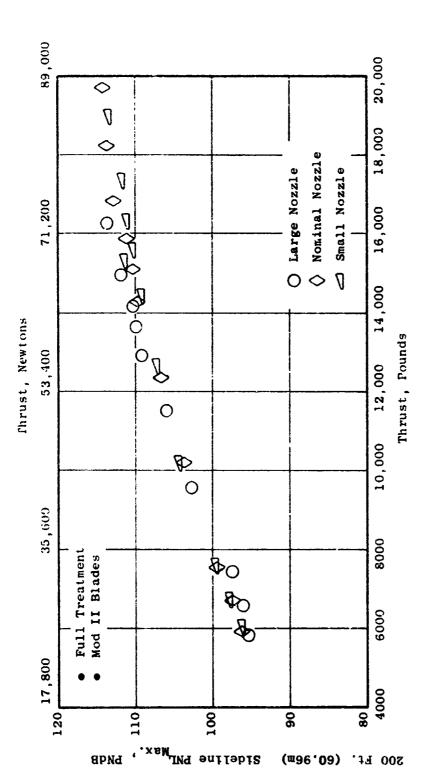


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200-ft (60.96 m) Sideline Aft Maximum PNL Vs. Thrust, Franc Treatment Configuration. Figure 18.



200-ft (60,96 m) Sideline Front Maximum PNL Vs. Thrust, Full Treatment Configuration. Figure 19.



200-ft (60.96 m) Sideline Aft Maximum PNL Vs. T.rust, Full Treatment Configuration. Figure 20.

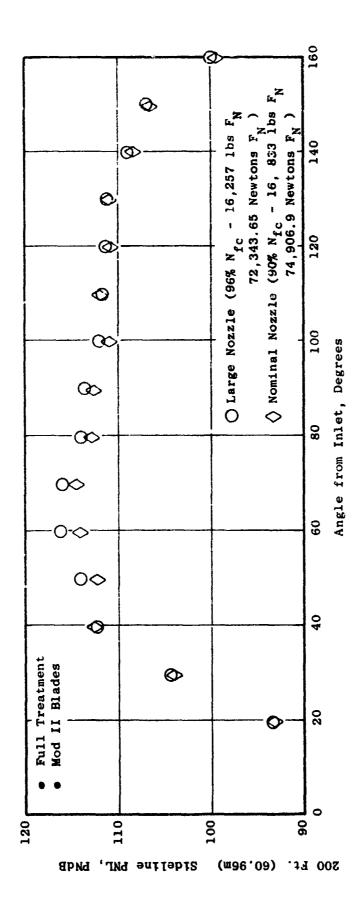


Figure 21, 200-ft (60,96 m) Sideline PML Vs. Angle from Inlet, Takeoff.

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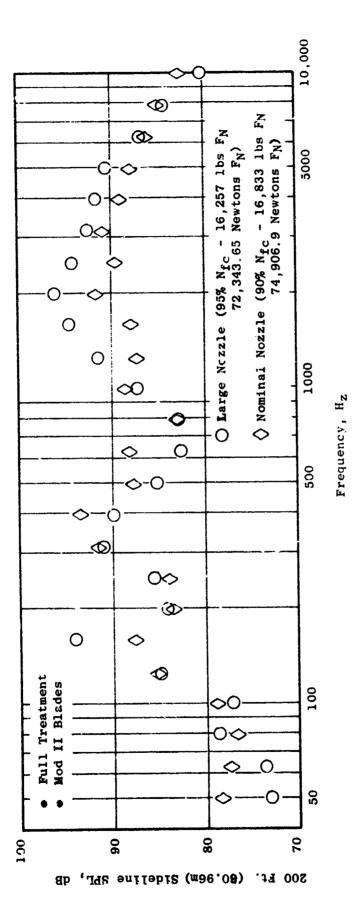


Figure 22, 200-ft (60,96 m) Sideline PNL Vs. Frequency, 60%.

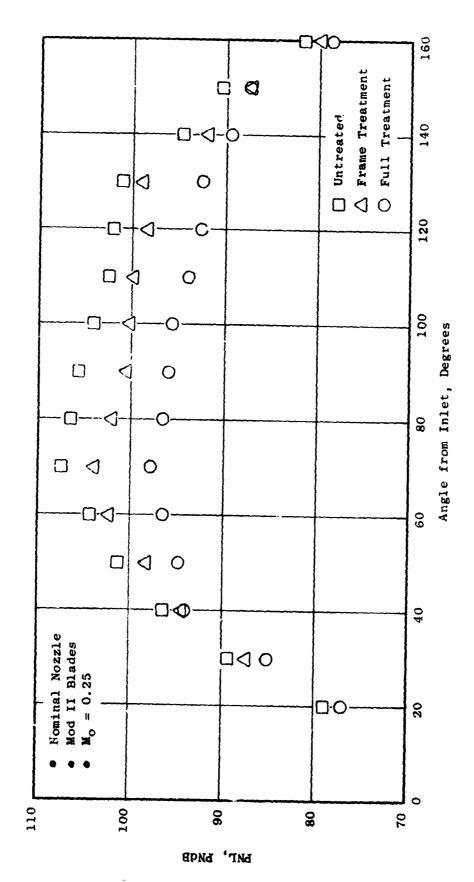


Figure 23. 1000-ft (304.8 m) Level Flyove PNL, Takeoff.

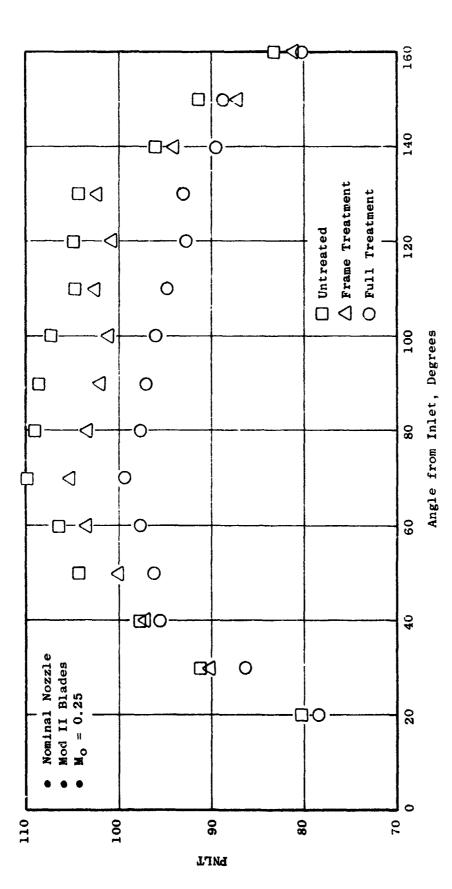


Figure 24. 1000-ft (304.8 m) Level Flyover PNLT, Takeoff.

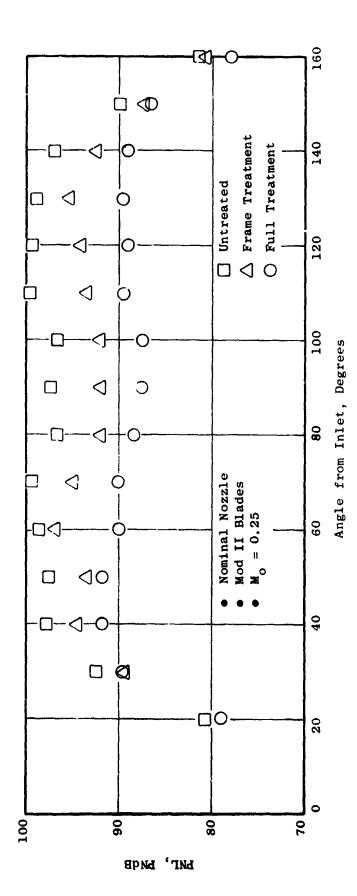


Figure 25. 370-ft (112.8 m) Level Flyover PML, Approacn.

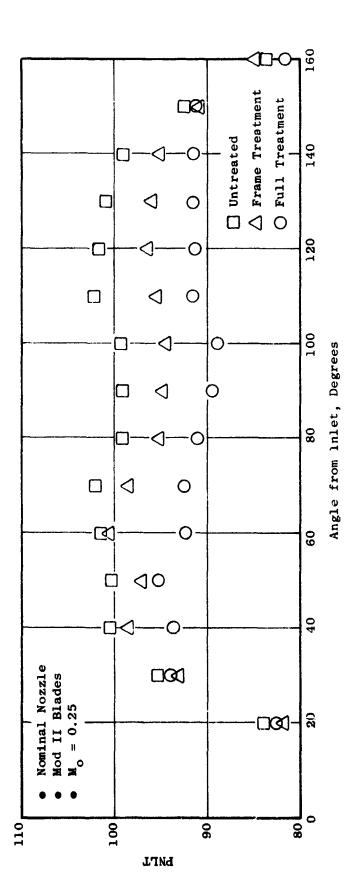
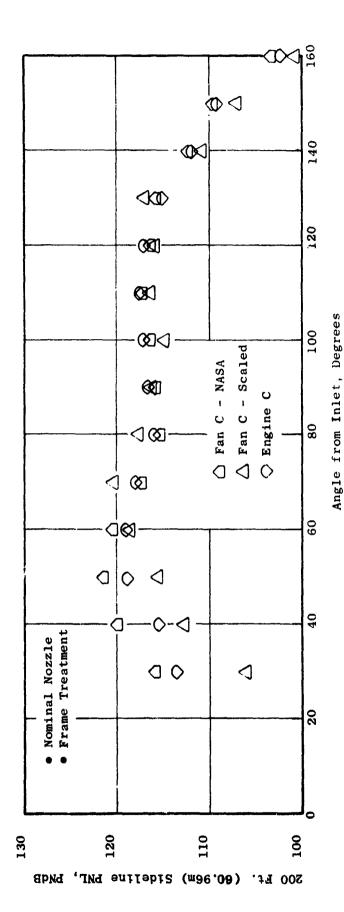
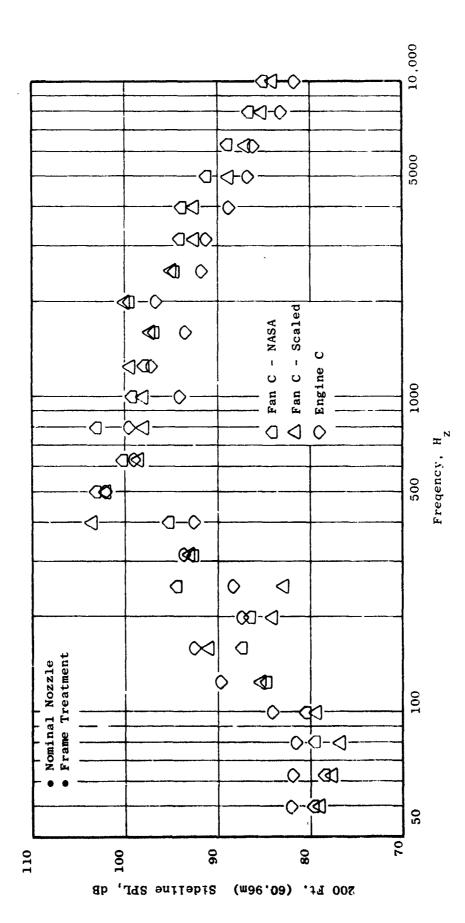


Figure 26, 370-ft (112,8 m) Level Flyover PMLT, Approach.

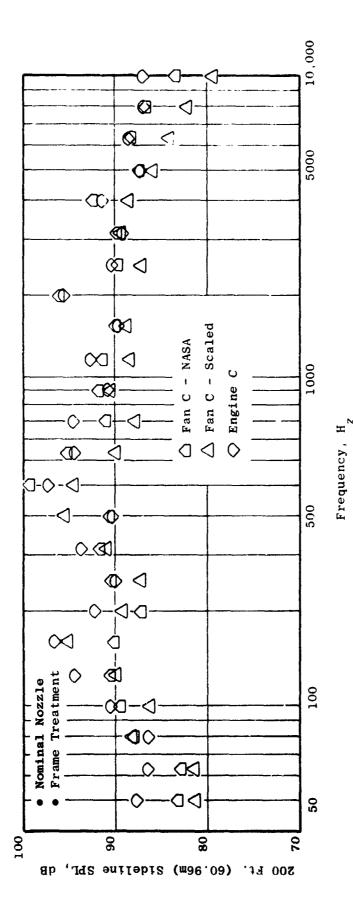
Figure 27. Asphalt to Gravel Surface Spectral Corrections.



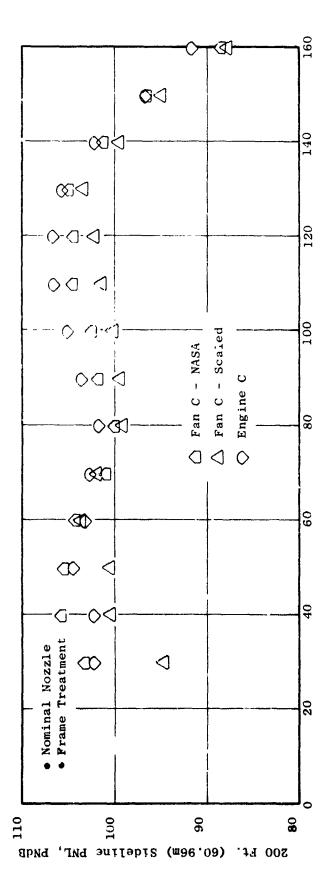
200-ft (60.96 m) Sideline PML Vs. Angle from Inlet for Full-Scale Fan, Scaled Fan, and Engine, Takeoff. Figure 28.



200-ft (60.96 m) Sideline SPI Vs. Frequency for Full-scale Fan, Scaled Fan, and Engine, 70°, Takeoff. Figure 29.

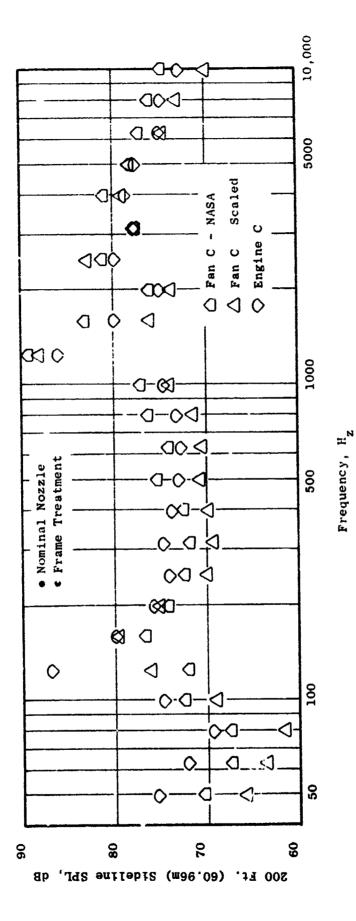


200-ft (60.96 m) Sideline SPL Vs. Frequency for Full-Scale Fan, Scaled Fun. and Engine, 110° , Takeoff.

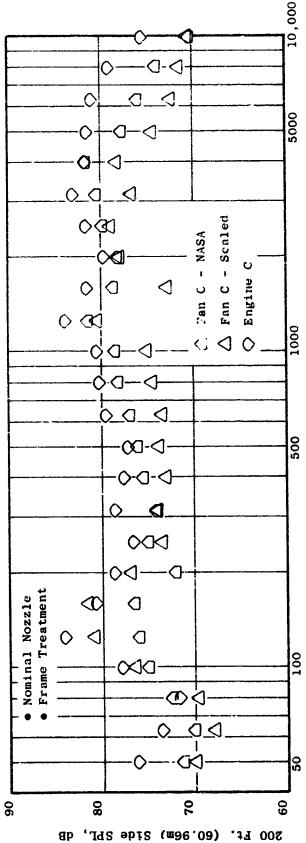


Angle from Inlet, Degrees

200-ft (60,96 m) Sideline PML Vs. Angle from Inlet for Full-Scale Fan. and Engine, Approach. Figure 31.



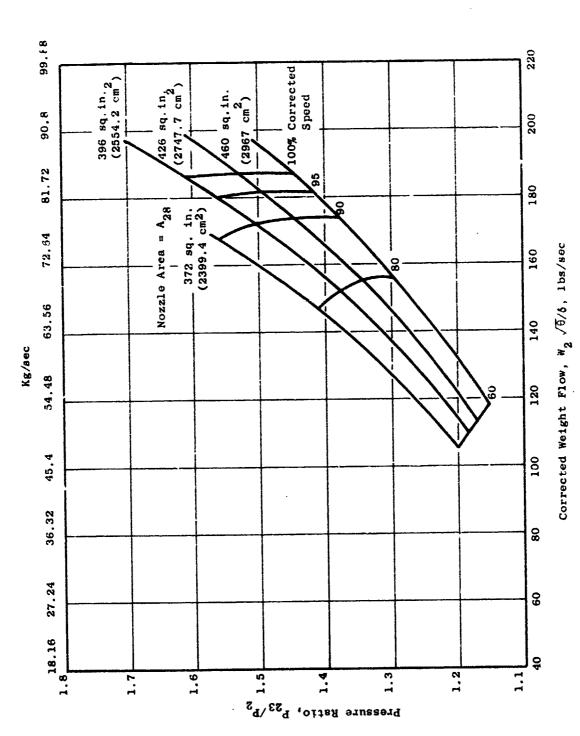
200-ft (60.96 m) Sideline SPL Vs. Frequency for Fill-Scale Fan, Scaled Fan, and Engine, 60°, Approach. Figure 32.



200-ft (60,96 m) Sideline SPL Vs. Frequency for Full-Scale Fan. Scaled Fan, and Engine, 120°, Approach. Figure 33.

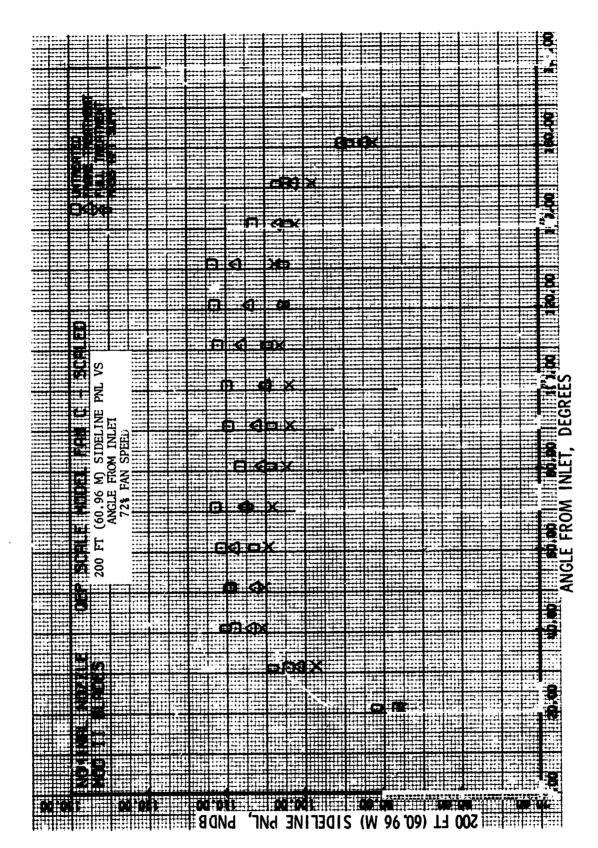
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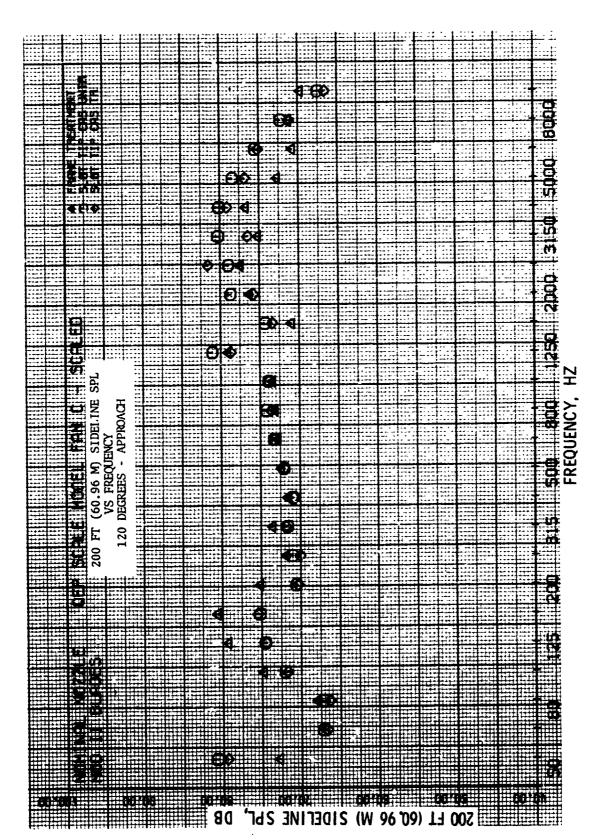


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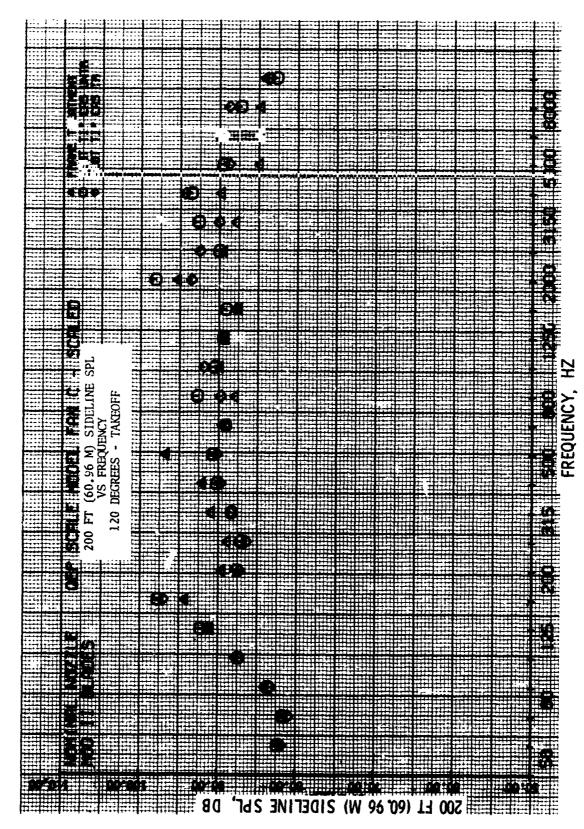
Figure 34. Scale Model Fan C Performance Map.



200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations, Takeoff. Figure 35.



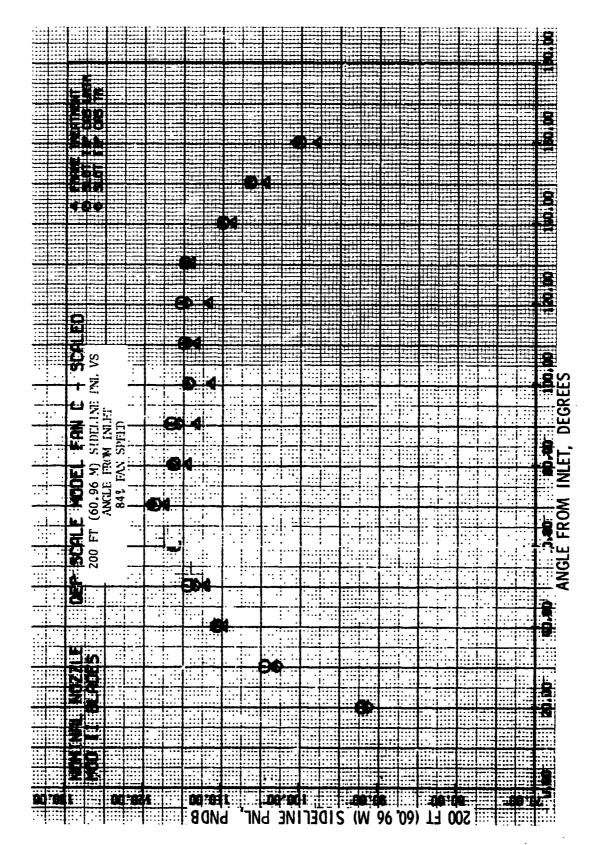
200-ft (60.96 m) Sideline SPL Vs. Frequency for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations, 70°, Takeoff. Figure 36.



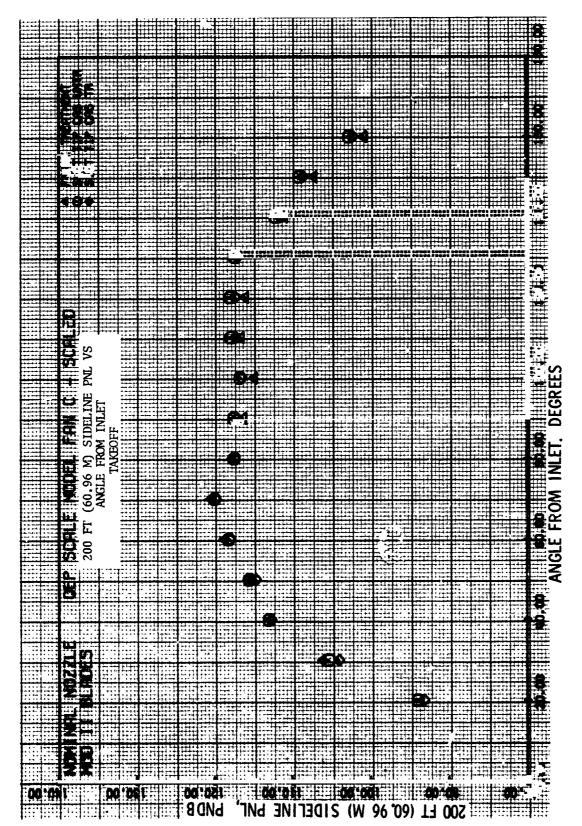
200-ft (60.96 m) Sideline SPL Vs. Frequency for Frame Treatment and Treated and Untrested Slotted Tip Casing Configurations, 120°, Takeoff. 37. Figure

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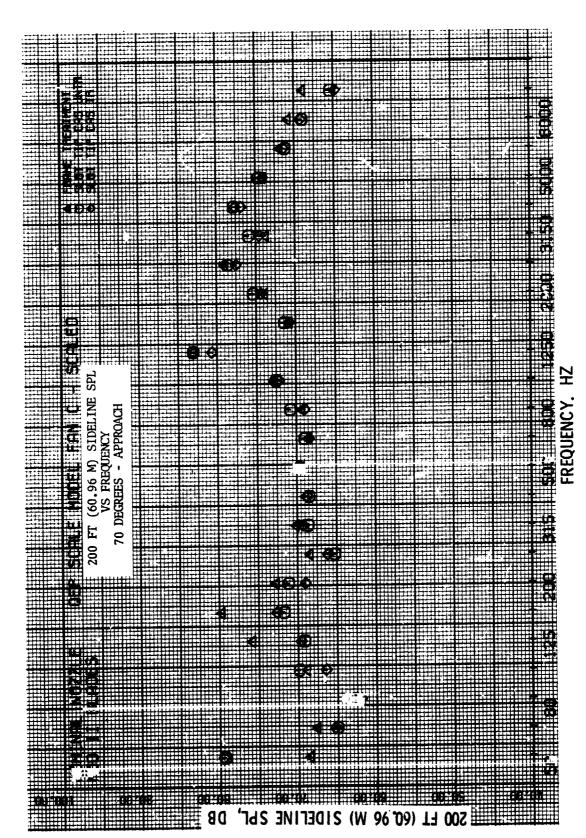
200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations, 84% Fan Speed. Figure 38.



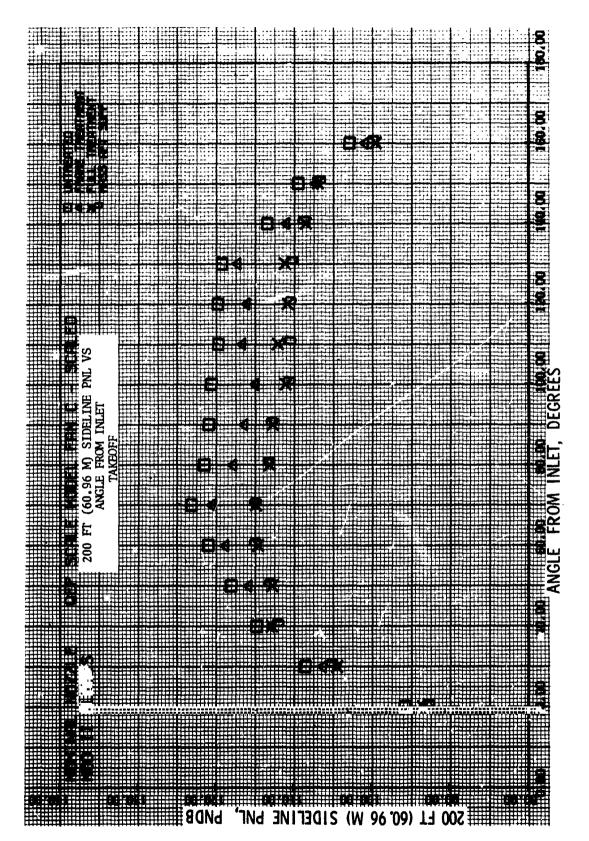
200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations, 72% Fan Speed. Figure 39.



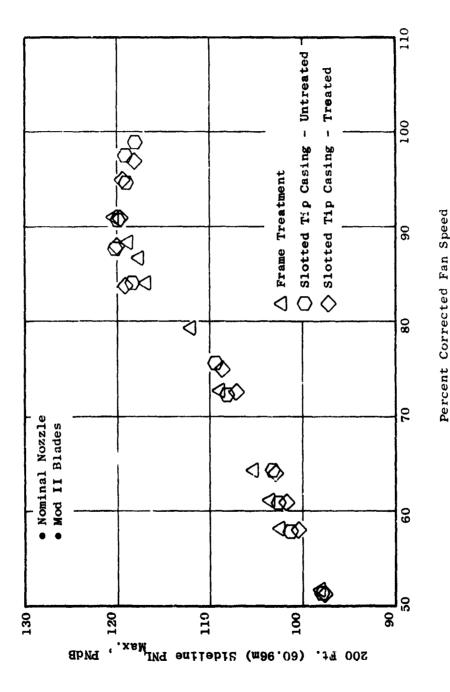
200-ft (60.96 m) Sideline PNL Vs. Angle from Inlet for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations, Approach. Figure 40.



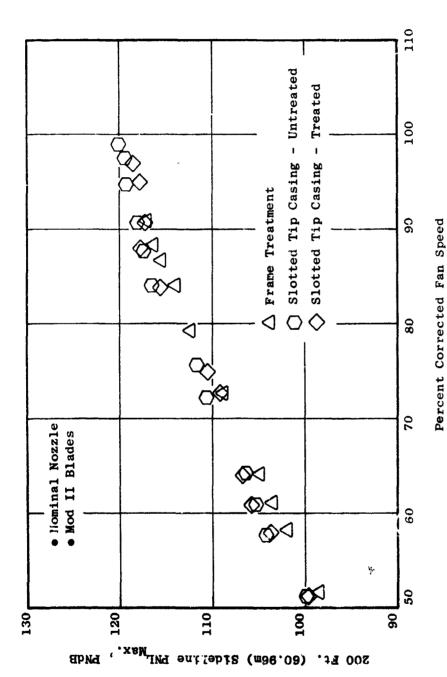
200-ft (60.96 m) Sideling SPL Vs. Frequency for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations, 70°, Approach. Figure 41.



200-ft (60.96 m) Sideline SPL Vs. Frequency for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations, 120°, Approach. 42. Figure

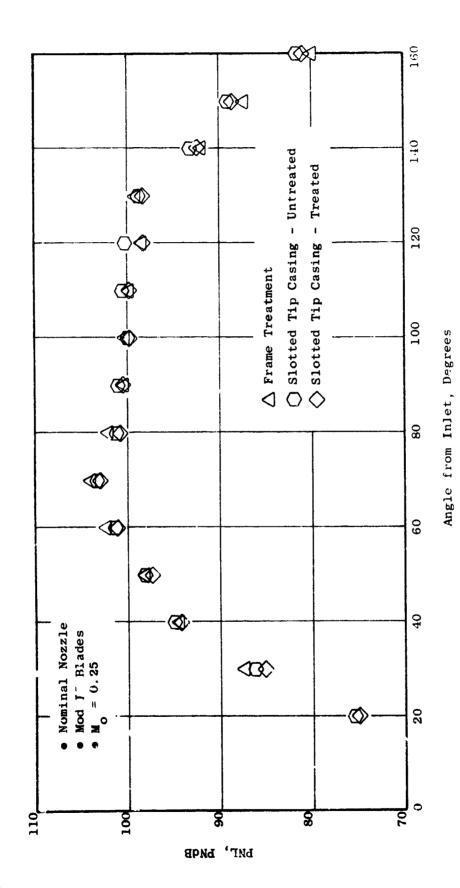


200-ft (60.96 m) Sideline Front Maximum PNL Vs. Corrected Fan Speed for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations. Figure 43.

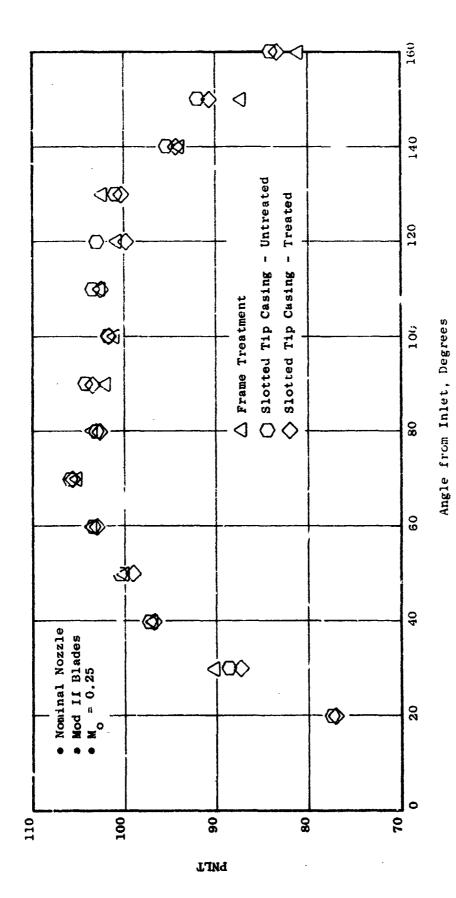


200-ft (60.96 m) Sideline Aft Maximum PNL Vs. Corrected Fan Speed for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations. Figure 44.

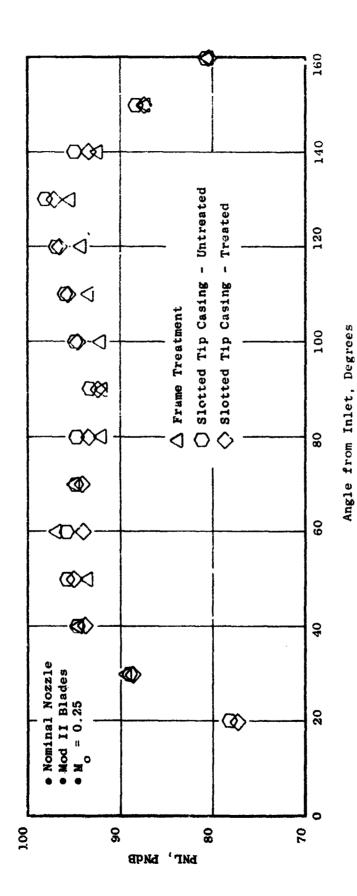
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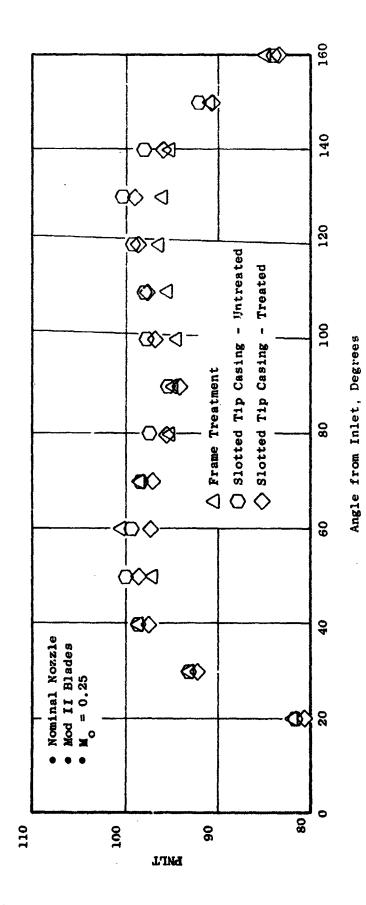
1000-ft (304,8 m) Level Flyover FNL for Frame Treatest's and Treatest and Universed Slotted Tip Casing Configurations, fakeoff. Figure 45.



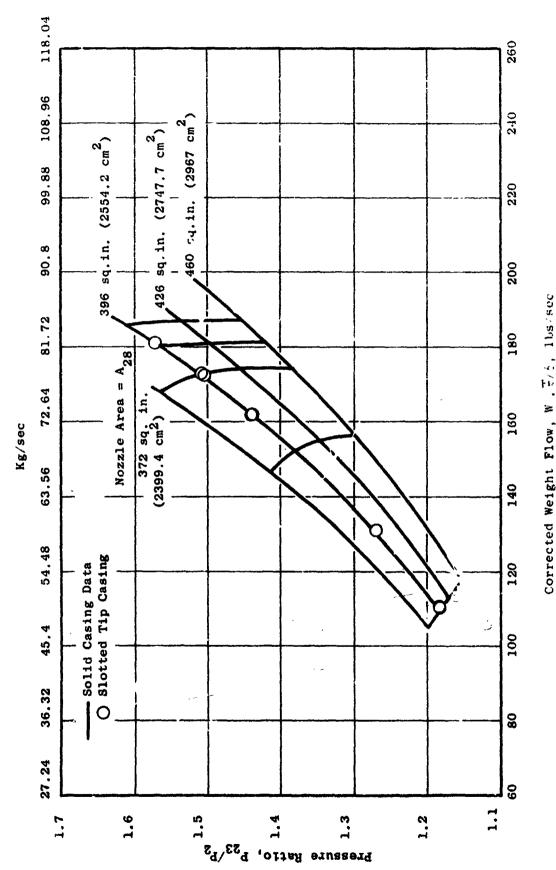
1000-ft (304.8 m) Level Flyover PNLT for Frame Treatment and Treated and Untreated . Takeo!f. Slotted Tip Casing Configuratio Figure 46.



370-ft (112.8 m) Level Flyover PNL for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations, Approach. Figure 47.

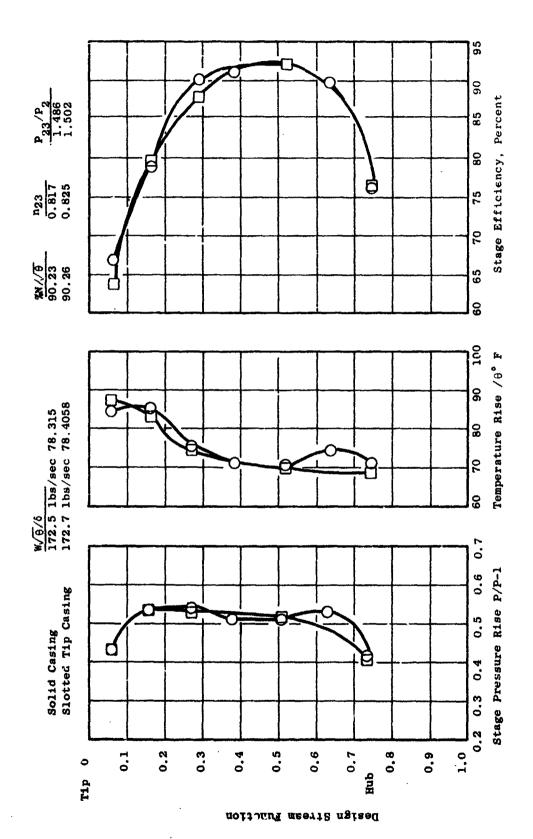


370-ft (112.8 m) Level Flyover PNLT for Frame Treatment and Treated and Untreated Slotted Tip Casing Configurations, Approach. Figure 48.



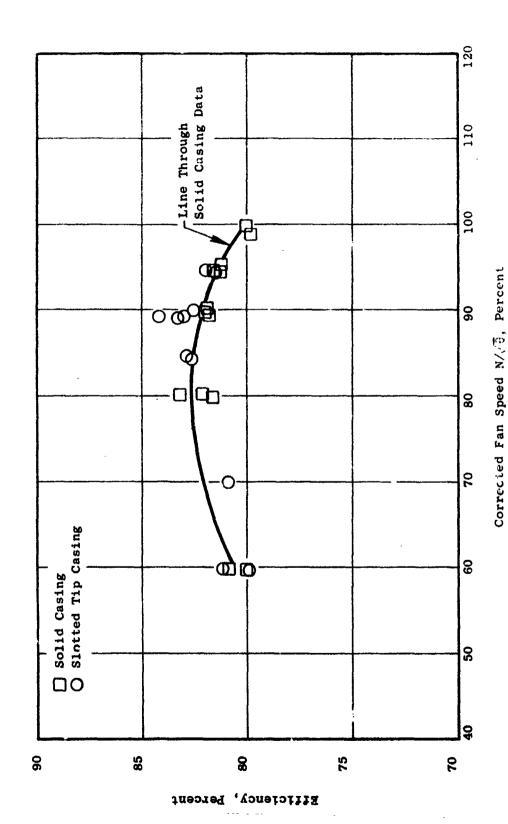
Performance Map showing Comparison of Solid Casing and Sletted Casing Data. Figure 49.

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Radial Distribution of Pressure Rise, Temperature Rise, and Besulting Efficiency at Takeoff Fan Speed. Figure 50.



Efficiency Vs. Corrected Fan Speed for Solid and Slotted Casings. Figure 51.

APPENDIX B One-Third Octave Data

The one-third octave data have been corrected to standard day (59° F, 70% R.H.). These data are on a 100-foot (30.48 m) arc for the scale model size and a 200-foot (60.96 m) sideline for the full-scale fan. A takeoff and approach set of data are included for the untreated; frame treatment, full treatment, slotted (untreated), and slotted (treated) configurations.

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UNTREATED NOWINAL NOZZLE APPROACH 200' (60.95 m) SIDELINE

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NOMINAL NOZZLE
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SLOTTED TIP CASING - TREATED NOMINAL NOZZLE TAKEOFF 200' (60.96 m) SIDELINE

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SLOTTED TIP CASING - TREATED NOMINAL NOZZLE APPROACH 200' (60.96 m) SIDELINE

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FULL TREATMENT
NOMINAL NOZZLE
APPROACH
100' (30.48 m) ARC
(SCALE MODEL DATA)

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FULL TREATMENT
NOMINAL NOZZLE
APPROACH
200' (60.96 m) SIDELINE
(SCALE MODEL - SCALED DATA)

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SLOTTED TIP CASING - UNTREATED NOMINAL NOZZLE TAKEOFF 100' (30.48 m) ARC (SCALE MODEL DATA)

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SLOTTED TIP CASING - UNTREATED
NOMINAL NOZZLE
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200' (G0.96 m) SIDELINE
(SCALE MODEL - SCALED DATA)

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APPROACH
100' (30.48 m) ARC
(SCALE MODEL DATA)

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SLOTTED TIP CASING - UNTREATED
NOMINAL NOZZLE
APPROACH
200' (60,96 m) SIDELINE
(SCALE MODEL - SCALED DATA)

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SLOTTED TIP CASING - TREATED NOMINAL NOZZLE

TAKEOFF

100' (30.48 m) ARC
(SCALE MODEL DATA)

(AND RADIANS) SLOTTED TIP CASING - TREATED NOMINAL NOZZLE

TAKEOFF

200 (60.96 m) SIDELINE
(SCALE MODEL - SCALED DATA)

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SLOTTED TIP CASING - TREATED NOMINAL NOZZLE

APPROACH 100' (30, 8 m) ARC (SCALE MODEL DATA) (AND RADIANS) TROM INLET IN DEGREES 90.0 OVERALL SECTO

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SLOTTED TIP CASING - TREATED NOMINAL NOZZLE
APPROACH
200' (60,96 m) SIDELINE

(SCALE MCJEL - SCALED DATA)

DAY

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Sendoria - America

APPENDIX C Nomenclature

BPF - blade passing frequency

EPNL - effective perceived noise level

F - degrees Fahrenheit

 F_{M} - corrected thrust, pounds

Hz - frequency, cycles per second

MPT - multiple pure tones

 N_{fc} - corrected fan speed

P₂₃'P₂ - fan pressure ratio

PNdB - PNL in decibels

PNL - perceived noise level

PNLT - tone corrected FNL

R.H. - celative humidity

SL - cideline

SPL - sound pressure level

TR - treated

UNTR - untreated

 W_2 - fan weight flow

the ratio of inlet pressure to standard pressure,
 14.7 pounds per square inch (10.13 Newtons/cm²)

n₂₃ - adiabatic efficiency

θ - temperature ratio

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